

AMENDMENTS

Amendments to the Specification:

Please replace page 1, lines 4-6, with the following amended lines:

The present invention relates to an improved assembly consisting of an internal casing and a support device for nozzles in a gas turbine stage.

Please replace page 4, lines 15-22, with the following amended lines:

The object of the present invention is therefore that of overcoming the drawbacks mentioned above and in particular that of providing an improved assembly consisting of an internal casing and a support device for nozzles in a gas turbine stage, which allows a reduction in the operating temperature of the components of the said assembly, with a consequent greater duration of said components.

Please replace page 4, lines 24-30, with the following amended lines:

Another object of the present invention is that of providing an improved assembly consisting of internal casing and support device for nozzles in a gas turbine state, which allows optimisation of the play between the rotor and stator of the turbine, with a consequent increase in the performance characteristics of the machine.

Please replace page 5, lines 7-11, with the following amended lines:

These and other objects according to the present invention are achieved by providing an improved assembly consisting of internal casing and support device for nozzles in a gas turbine stage, as illustrated in Claim 1.

Please replace page 7, lines 21-25, with the following amended lines:

These sector support devices or shrouds 114 are kept in position by the internal casing 112 which, with the aid of suitable grooves and pins 130 as well as by means of interlocking joints 116 with the said nozzles, prevents movement thereof.

Please replace page 7, lines 17-19, with the following amended lines:

Each nozzle sector is connected externally to the external casing 132 of the gas turbine by means of the support device 14 which is of the sector type and called a "shroud".

Please replace page 8, lines 1-9, with the following amended lines:

More precisely, the first cooling holes 122 of the internal casing 112 have an extension substantially parallel to the axis of the gas turbine. These holes have a diameter greater than that of the first holes 22 used in the assembly 10 known in the art, for example an approximate diameter of 1.8 mm. Advantageously a circumferential series of these first holes 22 may be provided, resulting for example in a total of forty-two first holes 22 for the entire internal casing 112.